COURSE CONTENT

COMPUTATIONAL STRUCTURE LABORATORY								
V SEMESTER :AE								
Course Code	Category	Hours / Week			Credits	Maximum Marks		
AAEC22	Core	L	T	P	C	CIA	SEE	Total
		0	0	3	1.5	30	70	100
Contact Classes: Nil	Tutorial Classes: Nil	Practical Classes: 36				Total Classes:36		
Prerequisite:								

I. COURSE OVERVIEW:

Computational Structure Laboratory course imparts detailed explanation on how to set up, run and interpret the results of various structures and covers all the necessary theoretical background for the industrial applications. Throughout the course ANSYS workbench will be used for all the structural analysis. This course offers a wide range of applications in aircraft structural analysis such as deflection of truss, frames, beams, stress and strain distributions in a plate as well as a solid continuum. It forms an essential cornerstone for mechanical, civil and aerospace engineers and plays a pivotal role in the efficient design of various structural systems.

II. COURSE OBJECTIVES:

The students will try to learn:

I. The structural analysis problems by defining proper boundary conditions, loads, and constraints, to accurately represent real-world scenarios.
II. The appropriate analysis method for different scenarios such as static analysis, dynamic analysis, linear and nonlinear analysis, and buckling analysis
III. The behavior of structures under different loading conditions, such as mechanical, thermal, or fluid loads using computational tools
IV. The visualization of simulation results, including stress distributions, deformation patterns, and safety factors.

III. COURSE OUTCOMES:

After successful completion of the course, students should be able to:

CO1 Utilize the static structural analysis of beams, truss, frame structure for assessing the structural behavior, integrity of system components.
CO2 Examine the dynamic behavior of spring mass system, simply supported beam for predicting natural frequency, mode shapes and resonance phenomenon in structural system.
CO3 Inspect the thermal behavior of bars and beams for the design and performance prediction of various structures in engineering applications.
CO4 Make use of nonlinear analysis for predicting failure modes and critical points in a system under nonlinear loading.
CO5 Identify the static, dynamic behavior of wing, fuselage and landing gear for deducing their structural integrity
CO6 Examine the static, dynamic behavior of composite materials under different loading and dynamic conditions for their suitability in aerospace applications.

IV.SYLLABUS

Exercises For Computational Structure Laboratory

Note: Students are encouraged to bring their own laptops for laboratory practice sessions.

1. Getting Started with Ansys Student Version

1.1 Ansys Student Version Installation procedure

System requirement

Supported Platforms and Operating Systems:

Microsoft Windows 10, 64-bit

Minimum Hardware Requirements for Ansys Student Product:

Processor(s): Workstation class

4 GB RAM

25 GB hard drive space

Computer must have a physical C:/" drive present

Graphics card and driver: Professional workstation class 3-D

OpenGL-capable

Installation Procedure

1. Extract (unzip) the downloaded installation files.
2. Right-click on setup.exe and select Run as Administrator. (This will run setup.exe from the extracted files.)
3. Read and accept the clickwrap to continue.
4. Click the right arrow button to accept the default values throughout the installation.
5. Click the exit button to close the installer.
6. The Ansys Student software is now installed.
7. Reboot your machine and then run the Ansys Student product from your Start menu by selecting Workbench.

Problem size limits

- No Geometry Export

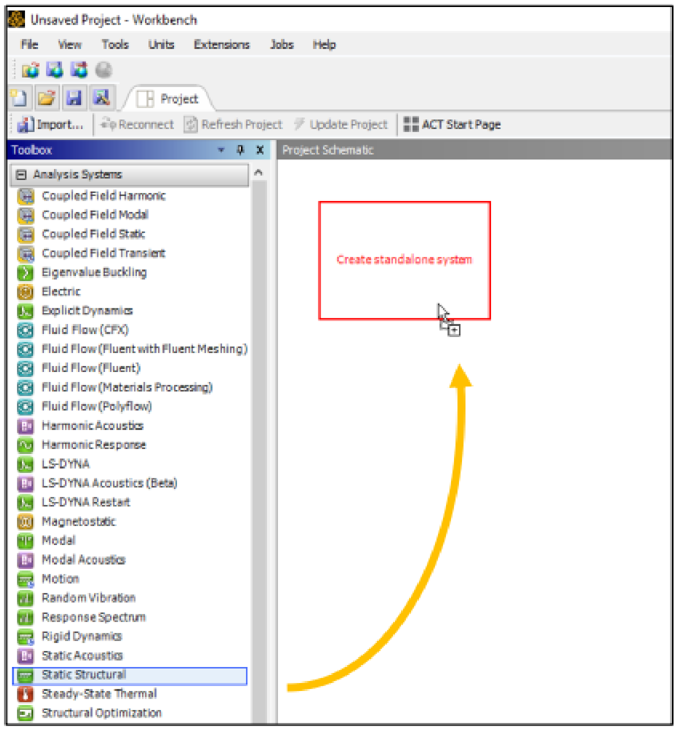
Limits for Ansys Student and Discovery (Refine Mode)

- Structural Physics: 128K nodes/elements
- Fluid physics: 1 Million cells/nodes

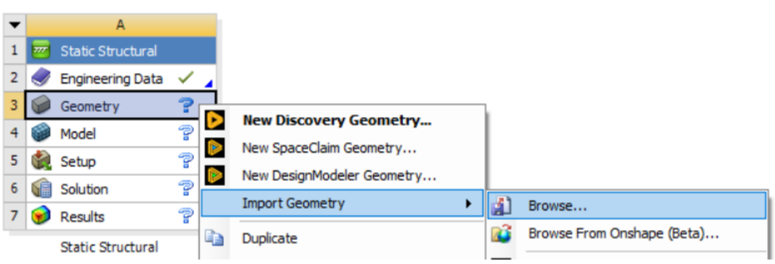
1.2 Getting Started with Ansys workbench

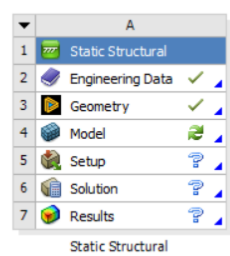
Open ANSYS Workbench: Windows Start Menu button → Ansys → Workbench • Under the Toolbox Analysis Systems category, click and drag analysis system Static Structural onto the Project Schematic,

drop it on target Create standalone system (this will be the only target available



• RMB—Geometry → Import Geometry → Browse… → valve_body_demo. scdoc → Open



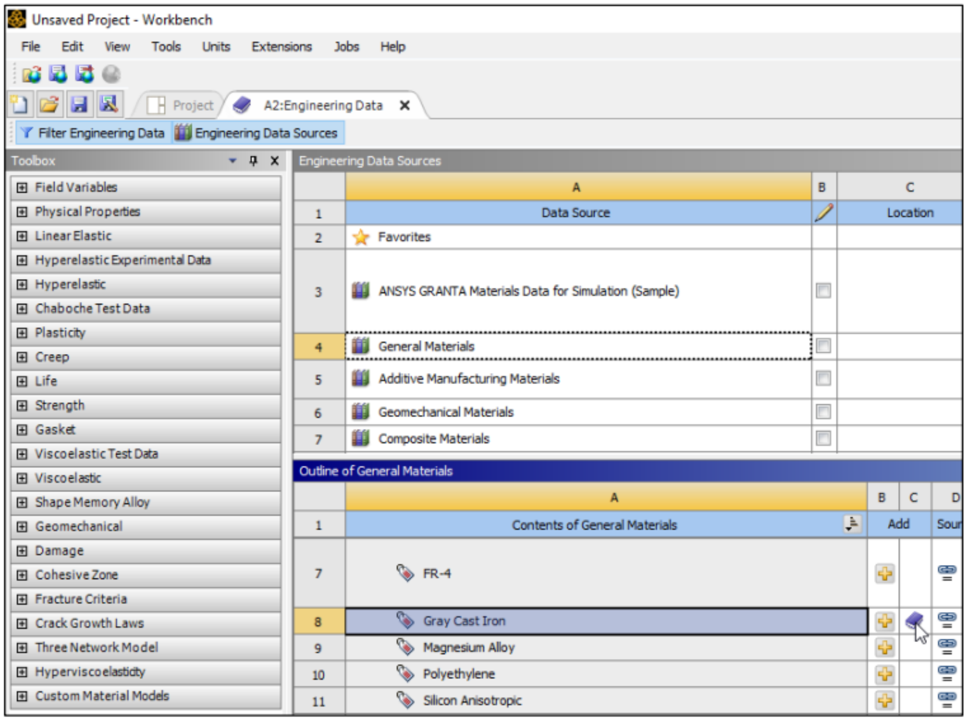


RMB—Engineering Data → Edit…

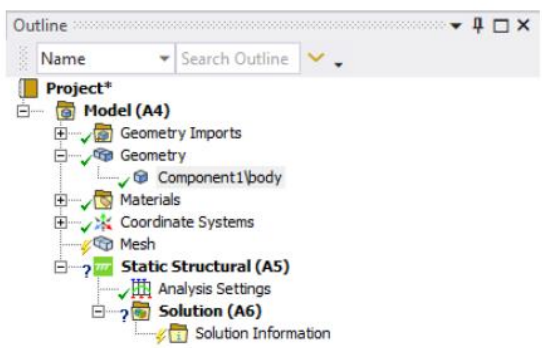
• Toggle toolbar button Engineering Data Sources to “on”

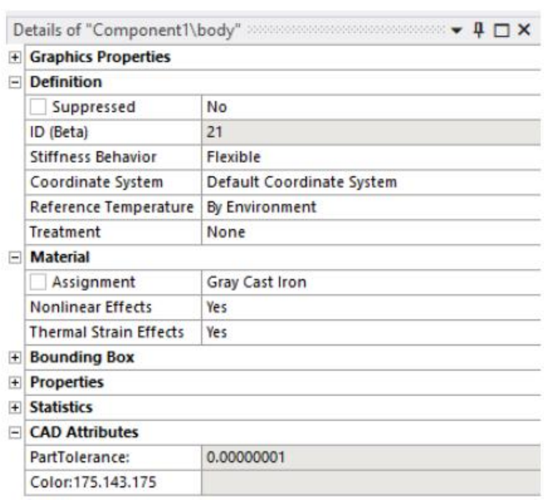
• Click Data Source General Materials

• In table Outline, click + icon next to Gray Cast Iron



- RMB—Model → Edit…
- Expand branch Geometry
- Select branch Component1\body
- In view Details, set Material →Assignment to Gray Cast Iron using drop-down menu





2. Exercises on Static analysis of Trusses

Static analysis is a fundamental engineering technique used to determine the forces, displacements, and stresses within structures under static loads. Trusses and frame structures are common examples of architectural and engineering designs that benefit from static analysis. Static analysis is crucial in designing trusses and frames that can safely support various loads while ensuring structural stability.

2.1 Static analysis (2D-Trusses)

Calculate the vertical deflection of the joint B and the horizontal movement of the support D in the truss shown in Fig. 2.1 The cross-sectional area of each member is 1800 mm2 and Young's modulus, E, for the material of the members is 200,000 N/mm2

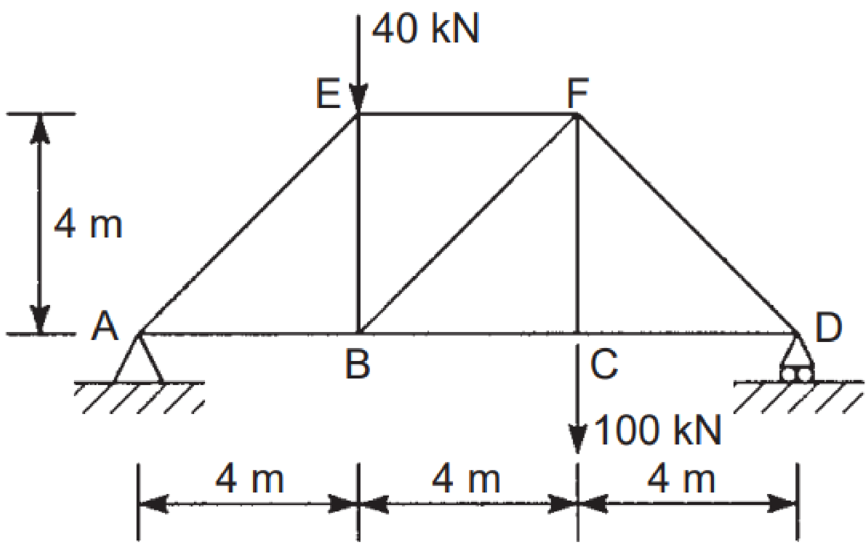


Fig. 2.1: 2DTruss with point loads

Hints

```
1. Pre-processor
```

```
• Element → Add → Link → 2D Spar → OK.
• Real constants → Add Set 1 → Cross section area → OK.
• Material Properties → Material Models → Structural → Linear → Elastic → Isotropic → Ex= 2.5e + 11 → PRXY=0.28 → OK.
• Modeling → Create → nodes → Inactive CS → enter node coordinates → OK → Elements → Auto numbered → Thru nodes → OK.
```

```
2. Solution
```

```
• Analysis Type → New analysis → Static → Ok
• Define loads → Apply → Structural → Displacement → On nodes → OK → All degrees
of freedom constrained → apply → Fx = – → OK
• Solve → Current LS → OK 3. General Post Processing
• Plot results → Deformed Shapes → Deformed + Undeformed
```

```
3. General Post Processing
```

```
• Plot results → Deformed Shapes → Deformed + Undeformed
```

2.1 Static analysis (3D-Trusses)

A three-dimensional truss has the cross sections of each of the truss members is 1.56e-3 sq meter. Assume the structure is made of aluminum with modulus of elasticity E=75 GPa. The structure is constrained in the X, Y and Z directions at the bottom three corners. The tower is loaded at the top tip. The load is in the YZ plane and makes an angle of 75 with the negative Y axis direction. The load value is 2500 N. Determine deflection at each joint, stress in each member and reaction forces at the base.

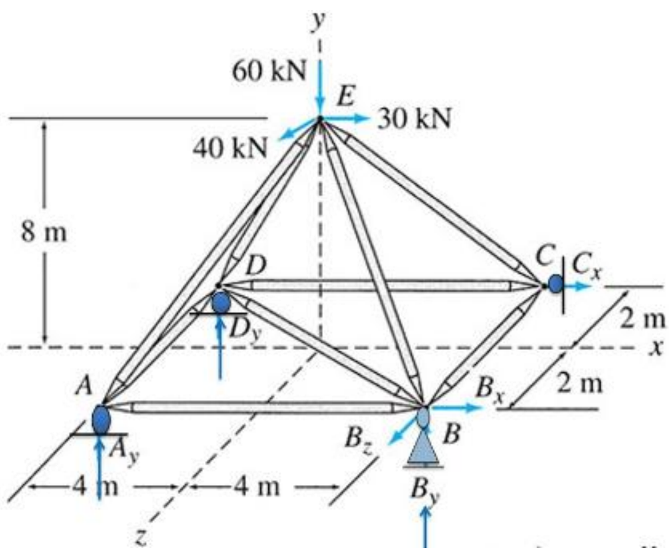


Fig. 2.2: 3DTruss with point loads

Hints

```
1. Pre-processor
```

```
• Element → Add → Link → 3D Spar → OK.
• Real constants → Add Set 1 → Cross section area → OK.
• Material Properties → Material Models → Structural → Linear → Elastic →
Isotropic → Ex= 2.5e + 11 → PRXY=0.28 → OK.
• Modeling → Create → nodes → Inactive CS → enter node coordinates → OK →
Elements → Auto numbered → Thru nodes → OK.
```

```
2. Solution
```

```
• Analysis Type → New analysis → Static → Ok
• Define loads → Apply → Structural → Displacement → On nodes → OK → All degrees
of freedom constrained → apply → Fx = -3.5 → OK
• Solve → Current LS → OK 3. General Post Processing
• Plot results → Deformed Shapes → Deformed + Undeformed ......
```

```
3. General Post Processing
```

```
• Plot results → Deformed Shapes → Deformed + Undeformed
```

Try

1. Modify the angle and magnitude of load in 2D truss and determine the deflection
2. Modify the angle and magnitude of load in 3D truss and deflection at each joint, stress in each member and reaction forces at the base

3. Exercises on Static analysis of beams

Static analysis of beams involves determining the internal forces, reactions, and deflections in a beam structure under various loading conditions. This analysis helps engineers design beams that can safely support applied loads.

3.1 Static analysis (Straight beam)

A cantilever beam made of mild steel is subjected to a point load as shown in Fig 3.1. Calculate the shear force, bending moment and stress distribution in the beam. L=150 cm, P=10kN, b=10cm, h=10cm

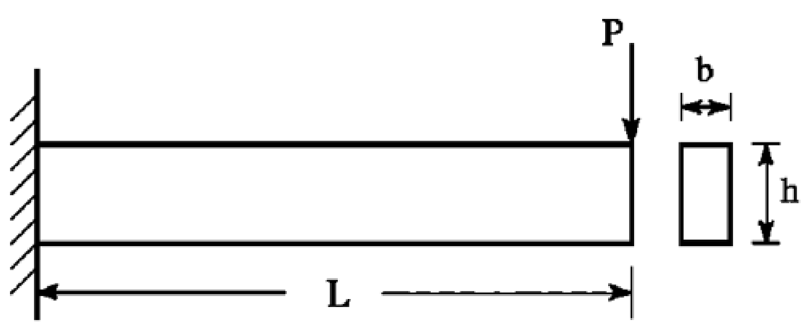


Fig. 3.1: Cantilever beam with point load at tip

Hints

```
1. Preferences: Structural Preprocessor:
```

```
• Element → Add → Beam3 • Real constants → Add Set1 → Cross section area = 100
cm^2 → Izz = 833.33 cm^4 → Height = 10 cm → Ok → Close
• Material Properties → Material Models → Structural → Linear → Elastic →
Isotropic → Ex = 2.5e7 → PRXY = 0.28 → Ok
• Modeling → Create → Key points → Inactive CS → (0,0,0) ;(100,0,0) ;(150,0,0)
→ Ok → Lines → Straight Lines → Ok
```

• Meshing → Size Controls → Manual Sizing → Lines → Picked Lines → No. of elements = 20 → Ok → Mesh → Lines → Ok

2. Solution:

• Analysis Type → New analysis → Static → Ok
• Define Loads → Apply → Structural → Displacement → On Key Points → Ok → All DOF Constrained → Ok → Force/Moments → On Key Points → FY = - 10000 → Ok
• Solve → Current LS → Ok

3.General Post Proc:

• Plot results → Deformed Shapes → Deformed + Un deformed
• Contour Plots → Nodal Solutions → DOF Solution → Y- Component Displacement Ok

3.2 Static analysis (Stepped beam)

A stepped cantilever beam made of mild steel is subjected to two pointed loads, one at the free end and one at the step as shown in figure below. Find the deflection and the displacement variation along the beam length. L=150cm, L1=100 cm, P1= P2=5kN, b1=12cm, b2=8cm, h1=12cm, h2=8cm.

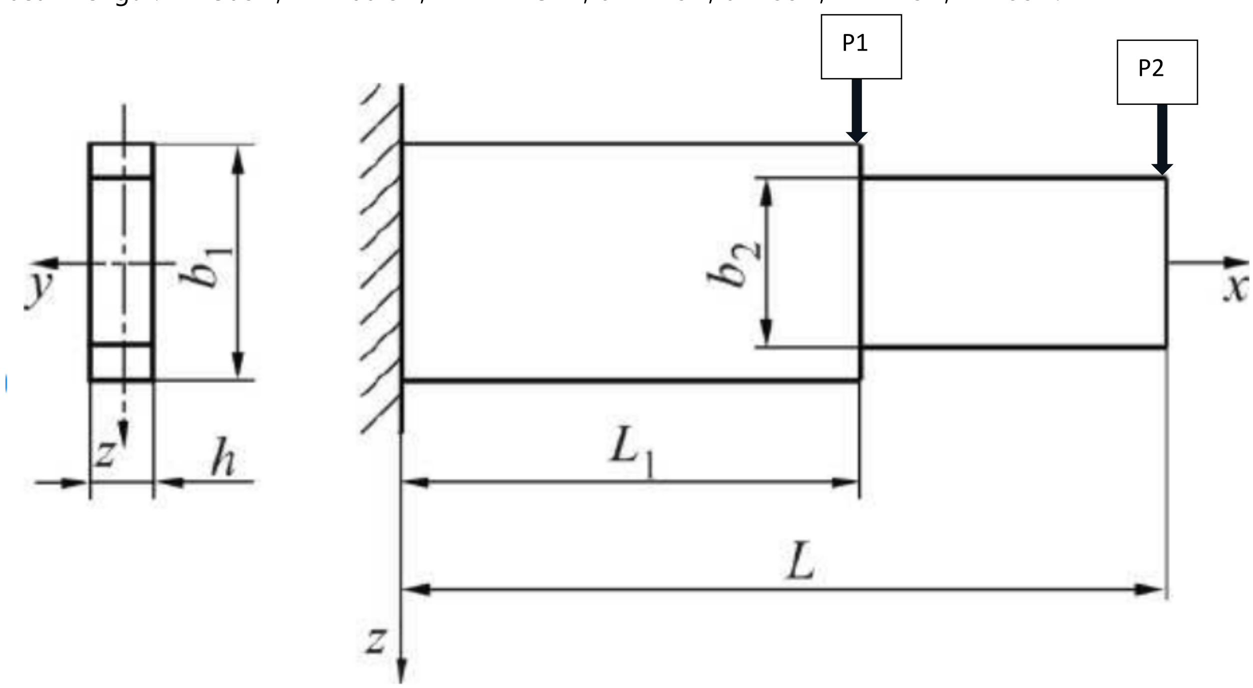


Fig. 3.2: Stepped Cantilever beam with point loads.

Hints

1. Preferences: Structural Preprocessor:

Element → Add → Beam3

• Real constants → Add Set1 → Cross section area = 144 cm^2 → Izz = 1728 cm^4 → Height = 12 cm → Ok → Add Set2 → Cross section area = 64 cm^2 → Izz = 341.33 cm^4 → Height = 8 cm → Ok → Close
• Material Properties → Material Models → Structural → Linear → Elastic → Isotropic → Ex = 2.5e7 → PRXY = 0.28 → Ok
• Modeling → Create → Key points → Inactive CS → (0,0,0) ;(100,0,0) ;(150,0,0) →Ok → Lines → Straight Lines → Ok
•Meshing → Mesh Attributes → Picked Lines → Ok → Real Constant set No.1 → Ok → Picked Lines → Ok → Real Constant set No.2 → Ok → Size Controls → Manual Sizing → Lines → Picked Lines → No. of elements = 10 → Ok → Mesh →Lines → Ok

2. Solution:

• Analysis Type → New analysis → Static → Ok
• Define Loads → Apply → Structural → Displacement → On Key Points → Ok → All DOF Constrained → Ok → Force/Moments → On Key Points → FY = - 5000 → Ok
• Solve → Current LS → Ok

3.General Post Proc:

• Plot results → Deformed Shapes → Deformed + Un deformed
• Contour Plots → Nodal Solutions → DOF Solution → Y- Component Displacement Ok

Try

1. Change the location of point load and calculate the shear force, bending moment and stress distribution in the beam

2. Change the location of point loads in tapered beam and find the deflection and the displacement variation along the beam length

4. Exercises on Static Analysis of Two-Dimensional Structures

4.1 Static analysis (Plate with a hole)

A plate with length of L=50mm and height of h=30 mm having hole at the center with radius r=15 mm is subjected to tensile load F= 100 N/mm2. The plate is made of aluminum with 3 mm thickness. Find the stress, strain and displacement.

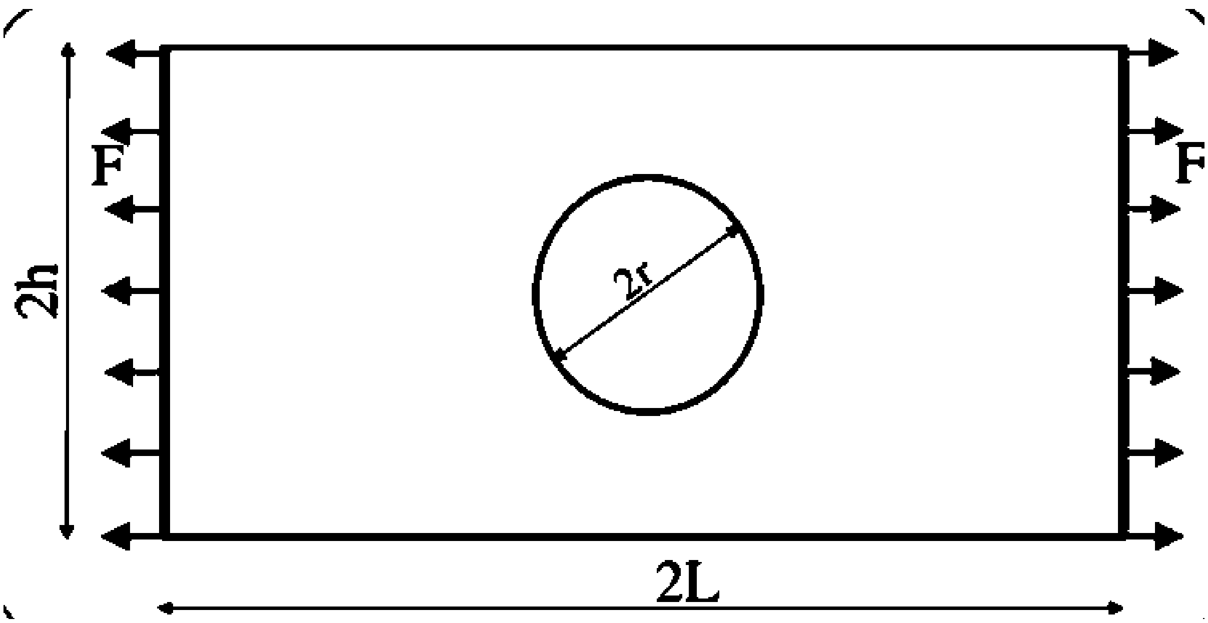


Fig. 3.2: Plate with hole

Hints

```
1.Preferences: Structural Preprocessor:
• Element → Add → Solid → Quad 4 node 42 → Ok → Options → Element Behavior = Plane stress w/thk → Ok
• Real constants → Add Set1 → THK = 3 mm → Ok → Close
• Material Properties → Material Models → Structural → Linear → Elastic → Isotropic → Ex = 0.7e5 → PRXY = 0.32 → Ok
• Modeling → Create → Areas → Rectangle → By Centre and Corner → PX = 0,PY = 0, Width = 50, Height = 30 → Ok → Circle → Two end pts →(-10,0),(10,0) → Ok
• Operate → Booleans → Subtract → Select Area → Ok
• Meshing → Size Controls → Manual Sizing → Areas → All Areas → Element Edge Length = 5 → Ok → Mesh → Areas → Free → Ok


Solution:
• Analysis Type → New analysis → Static → Ok • Sol’n Controls → Automatic Time Stepping = On → No. of Sub-steps = 5 → Max. No. of Sub-steps = 1000 → Min. No of Sub-steps = 1 → Ok
```

```
• Define Loads → Apply → Structural → Displacement → On lines → Ok → All DOF
Constrained → Ok → Pressure → On lines → 100 N/(mm^2) → Ok
• Solve → Current LS → Ok
General Post Processing:
   • Plot results → Deformed Shapes → Deformed + Un deformed
   • Contour Plots → Nodal Solution → Stress → X Comp. → Ok
   • Contour Plots → Nodal Solution → Stress → XY Shear → Ok
   • Contour Plots → Nodal Solution → Stress → Von Mises Stress → Ok Results:
```

Try

1. Change the number of holes, location and find the maximum stress concentration, strain and displacement.
2. Fix the left side of plate and apply compressive loads on other three sides and find the maximum stress concentration, strain and displacement.

5. Exercises on Dynamic Analysis: Modal and Transient Analysis

5.1 Dynamic Analysis (Model analysis of beam)

A cantilever beam made of mild steel as following specifications L=150 cm, b=10cm, h=10cm shown in Fig 5.1. Calculate the natural frequency of beam and compare it with the analytical results.

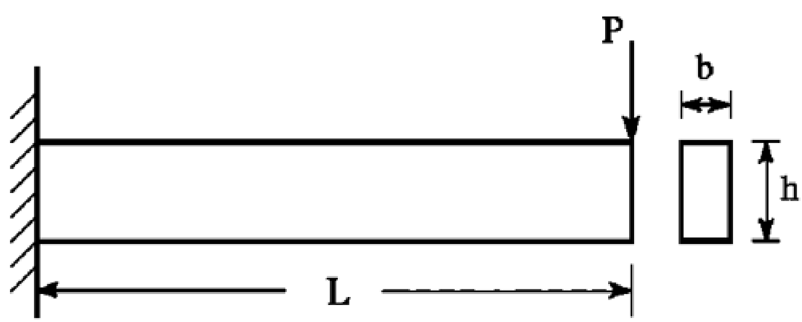


Fig. 5.1: Cantilever beam

Hints

```
1.  Preferences: Structural Preprocessor:
```

```
• Element → Add → Beam3 21
• Real constants → Add Set1 → Cross section area = 100 cm^2 → Izz = 833.33 cm^4 →
Height = 10 cm → Ok → Close
• Material Properties → Material Models → Structural → Linear → Elastic → Isotropic →
Ex = 2.5e7 → PRXY = 0.28 → Density = 7800kg/(m^3) →Ok
• Modeling → Create → Key points → Inactive CS → (0,0,0) ;(100,0,0) ;(150,0,0) → Ok →
Lines →Straight Lines → Ok
• Meshing → Size Controls → Manual Sizing → Lines → Picked Lines → No. of elements =
10 → Ok → Mesh → Lines → Ok 6.4.2
```

```
2. Solution:
```

```
• Analysis Type → New analysis → Modal → Ok
• Analysis Option → Block Lancoz → No. of Modes to extract = 5 → No. of Modes to expand = 5 → Ok
• Solve → Current LS → Ok
General Post Proc:
• Plot results → Deformed Shapes → Deformed + Un deformed
```

Try

1. Use the aluminum as material and find the natural frequency for same beam.
2. Take a simply supported beam made of mild steel and find the natural frequency.

6. Exercises on Thermal Analysis

6.1. Thermal Analysis (Steady State conduction through plate)

A square plate shown in Fig 6.1 has thermal conductivity of 10 W/m^{0}C. The top side is maintained at 250^{0}C and other three sides are maintained at 40^{0}C. Determine temperature at the center of the plate and temperature distribution from top to bottom passing through the center.

Hints

```
1. Preferences: Structural Preprocessor:
```

```
Preprocessor → Modeling → Create → Areas → Rectangle → By 2 Corners → X=0, Y=0, Width=1, Height=1

BLC4,0,0,1,1
Preprocessor → Element Type → Add/Edit/Delete... → click 'Add' → Select Thermal Mass Solid, Quad 4Node 55 ET,1, PLANE55

Preprocessor → Material Props → Material Models → Thermal → Conductivity → Isotropic → KXX = 10 (Thermal conductivity) MP, KXX,1,10

Preprocessor → Meshing → Size Cntrls → ManualSize → Areas → All Areas → 0.05 AESIZE,ALL,0.05

Preprocessor → Meshing → Mesh → Areas → Free → Pick All AMESH, ALL
```

```
• Solve → Current LS → Ok
General Post Proc:
• Plot results → Temperature distribution
```

Try

1. Instead of temperature apply heat flux on the sides of square and find the temperature distribution.
2. Introduce a hole at the center of square and find the temperature distribution

7. Exercises on Nonlinear Analysis

7.1 Nonlinear Analysis (Cantilever beam)

A cantilever beam with length of 550 cm and cross section area of 100 mm2 subjected to nonlinear load.. Find the deflection and nonlinear behavior.

Hints

```
1.  Preferences: Structural Preprocessor:
```

```
• Element → Add → Beam 3
• Material Properties → Material Models → Structural → Linear → Elastic → Isotropic → Ex = 2e11 → PRXY = 0.33 → Density = 7850kg/(m^3) →Ok
• Modeling → Create → Key points → Inactive CS → (0,0,0); (100,0,0) → Ok → Lines → Areas → Ok
• Meshing → Size Controls → Manual Sizing → Lines → Picked Lines → No. of elements = 20 → Ok → Mesh → Lines → Ok
• Select parameters → select functions define
```

```
2. Solution:
```

```
• Analysis Type → New analysis →Transient → Ok
• Parameters →functions→ define/edit→ type in result
• Select file→ file name= transient→ desktop→ save
• Parameters→ functions→ read from file→ open transient→ give table parameter name cantilever
• Select loads→ define loads→ apply→ structural→ displacement→ on keypoints→ all DOF→ keypoint 1→ ok
```

Try

1. Use a cantilever beam with same length as stated in above problem but with circular cross section and find its nonlinear behavior.
2. Use the same cantilever beam subjected to torque and find its nonlinear behavior.

---

8. Exercises on Harmonic Response Analysis

8.1 Harmonic Response Analysis (Cantilever beam)

A cantilever beam made of mild steel as following specifications L=150 cm, b=10cm, h=10cm shown in Fig 8.1.is subjected to a periodic force, which is mathematically represented below. The amplitude of the force is 1000 N. P = 1000k sin $(\frac{\pi t}{4})$. Find the deflection of beam.

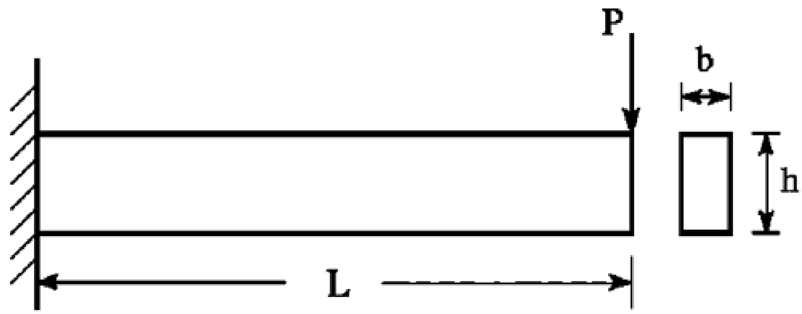


Fig 8.1. Cantilever beam with periodic force

Hints

```
1.      Preferences: Structural Preprocessor:
```

```
• Element → Add → Beam 3
• Material Properties → Material Models → Structural → Linear → Elastic →
Isotropic → Ex = 2e11 → PRXY = 0.33 → Density = 7850kg/(m^3) →Ok
• Modeling → Create → Key points → Inactive CS → (0,0,0); (100,0,0) → Ok →
Lines → Areas → Ok
• Meshing → Size Controls → Manual Sizing → Lines → Picked Lines → No. of
elements = 20 → Ok → Mesh → Lines → Ok
• Select parameters → select functions define
```

```
3. Solution:
```

```
• Analysis Type → New analysis → → Ok
• Parameters →functions→ define/edit→ type in result 1000k sin ( πt/4)
• Select file→ file name= transient→ desktop→ save
• Parameters→ functions→ read from file→ open transient→ give table parameter
name cantilever
• Select loads→ define loads→ apply→ structural→ displacement→ on keypoints→
all DOF→ keypoint 1→ ok
• Solve → Current LS → Ok
```

Try

1. Repeat the above analysis for simply supported beam.
2. Change the magnitude of periodic load and repeat the same analysis.

9. Exercises on Analysis of Aircraft Structure:Wing

9.1 Static structural (Wing)

Perform static structural analysis of an aircraft rectangular wing of a span 100 cm and the chord length of wing is 10 cm and cross-sectional area is 100 cm2. Find the deflection and deformation of wing.

```
1. Preferences: Structural Preprocessor:
```

• Element → Add → wing • Real constants → Add Set1 → Cross section area = 100 cm^2 → Izz = 833.33 cm^4 → Height = 10 cm → Ok → Close
• Material Properties → Material Models → Structural → Linear → Elastic → Isotropic → Ex = 2.5e7 → PRXY = 0.28 → Ok
• Modeling → Create → Key points → Inactive CS → (0,0,0) ;(100,0,0) ;(150,0,0) → Ok → Lines → Straight Lines → Ok
• Meshing → Size Controls → Manual Sizing → Lines → Picked Lines → No. of elements = 20 → Ok → Mesh → Lines → Ok

2. Solution:

• Analysis Type → New analysis → Static → Ok
• Define Loads → Apply → Structural → Displacement → On Key Points → Ok → All DOF Constrained → Ok → Force/Moments → On Key Points → FY = - 10000 → Ok
• Solve → Current LS → Ok

3.General Post Proc:

• Plot results → Deformed Shapes → Deformed + Un deformed
• Contour Plots → Nodal Solutions → DOF Solution → Y- Component Displacement Ok

Try

1. Repeat the above analysis for a tapered wing.
2. Repeat the above analysis for tapered wing with uniformly varying load

9.2 Dynamic Analysis (Model analysis of wing)

Perform model analysis of an aircraft rectangular wing of a span 100 cm and the chord length of wing is 10 cm and cross-sectional area is 100 cm2. Calculate the natural frequency of wing.

Hints

1. Preferences: Structural Preprocessor:

• Element → Add → wing
• Real constants → Add Set1 → Cross section area = 100 cm^2 → Izz = 833.33 cm^4 → Height = 10 cm → Ok → Close
• Material Properties → Material Models → Structural → Linear → Elastic → Isotropic → Ex = 2.5e7 → PRXY = 0.28 → Density = 7800kg/(m^3) →Ok
• Modeling → Create → Key points → Inactive CS → (0,0,0) ;(100,0,0) ;(150,0,0) → Ok → Lines →Straight Lines → Ok
• Meshing → Size Controls → Manual Sizing → Lines → Picked Lines → No. of elements = 10 → Ok → Mesh → Lines → Ok 6.4.2

2. Solution:

• Analysis Type → New analysis → Modal → Ok
• Analysis Option → Block Lancoz → No. of Modes to extract = 5 → No. of Modes to expand = 5 → Ok
• Solve → Current LS → Ok

Try

1. Repeat the above analysis for a tapered wing.
2. Repeat the above analysis for tapered wing with uniformly varying load

10. Exercises on Analysis of Aircraft Structure: Fuselage

10.1 Static structural (Fuselage)

Calculate the deformation of the aluminum fuselage section under the application of internal load of 100000 Pa. The radius of fuselage is 0.15m and thickness is 2 mm.

1. Preferences: Structural Preprocessor:

```
•Element type → Add / edit/Delete → Add → Solid – 10 node 92→ Apply Add → Beam 2 Node 188 → Apply → Add → Shell →Elastic 4 node 63 Real Constants → Add → Select shell → give thickness (I) = 1→ ok → close.
•Material properties → material models → Structural → Linear → Elastic → Isotropic EX = 0.7e11; PRXY = 0.3; Density = 2700
•Pre-processor → modelling → Create → Areas → Circle → Annulus WP x = 0 ; WP y = 0; Rad – 1 = 2.5; Rad - 2 = 2.3 OK Pre-processor → Modelling → Create → Circle → Solid – WP x = 0; X = 2.25; Y = 0 Radius = 0.15 Apply WP x = 0; X = -2.25; Y = 0 Radius = 0.15 Apply WP x = 0; X =0; Y = 2.25; Radius = 0.15 Apply WP x = 0; X = 0; Y = -2.25 Radius = 0.15 OK
•Pre-processor → Modelling → Operate → Booleans → Add → Areas – Pick all OK
•Preprocessor → Modelling → Operate → Extrude → Areas → By XYZ offset X= 0; Y=0; Z = 5
•Pre-processor →Meshing → Size controls → Manual Size → All Areas → give element edge length as 0.15 → ok
• Meshing → Size controls → Manual Size → All lines → give element edge length as→ ok
•Meshing → Mesh → areas → free → select box type instead of single → select the total volume → ok
```

2. Solution:

```
•Loads → define loads → Apply → Structural → Displacement → On areas → select box type → select box (4 points at centre) → all DOF → ok Select → ALL DOF arrested Define loads → Apply → Structural → Pressure → on areas → select the internal surface of the fuselage and give value (100000) → ok
• Solve → Current LS → Ok
General Post Proc:
• Plot results → Deformed Shapes → Deformed + Un deformed
```

Try

1. Repeat the above analysis for a monocoque fuselage.
2. Repeat the above analysis for a monocoque fuselage with both internal and external load.

10.2 Dynamic analysis (Model analysis of Fuselage)

Calculate the natural frequency of the aluminum fuselage section under the application of internal load of 100000 Pa. The radius of fuselage is 0.15m and thickness is 2 mm.

1. Preferences: Structural Preprocessor:

```
•Element type → Add / edit/Delete → Add → Solid – 10 node 92→ Apply Add → Beam 2 Node 188 → Apply → Add → Shell →Elastic 4 node 63 Real Constants → Add → Select shell → give thickness (I) = 1→ ok → close.
```

```
•Material properties → material models → Structural → Linear → Elastic → Isotropic EX = 0.7e11; PRXY = 0.3; Density = 2700
•Pre-processor → modelling → Create → Areas → Circle → Annulus WP x = 0 ; WP y = 0; Rad – 1 = 2.5; Rad - 2 = 2.3 OK Pre-processor → Modelling → Create → Circle → Solid – WP x = 0; X = 2.25; Y = 0 Radius = 0.15 Apply WP x = 0; X = -2.25; Y = 0 Radius = 0.15 Apply WP x = 0; X =0; Y = 2.25; Radius = 0.15 Apply WP x = 0; X = 0; Y = -2.25 Radius = 0.15 OK
•Pre-processor → Modelling → Operate → Booleans → Add → Areas – Pick all OK
•Preprocessor → Modelling → Operate → Extrude → Areas → By XYZ offset X= 0; Y=0; Z = 5
•Pre-processor →Meshing → Size controls → Manual Size → All Areas → give element edge length as 0.15 → ok
• Meshing → Size controls → Manual Size → All lines → give element edge length as→ ok
•Meshing → Mesh → areas → free → select box type instead of single → select the total volume → ok
```

```
2. Solution:
```

```
• Analysis Type → New analysis → Modal → Ok
• Analysis Option → Block Lancoz → No. of Modes to extract = 5 → No. of Modes to expand = 5 → Ok
• Solve → Current LS → Ok
```

Try

1. Repeat the above analysis for a monocoque fuselage.
2. Repeat the above analysis for a monocoque fuselage with both internal and external load.

11. Exercises on Analysis of Aircraft Structure: Landing Gear

11.1 Static structural (Landing gear)

A simple retractable landing gear subjected to a load of 10000 N. Find the deformation and stress developed in the landing gear.

```
1. Preferences: Structural Preprocessor:
```

```
•Preferences → Structural → H-Method → OK
•Preprocessor → Element Type → Add → Add → Select Link → 2D spar 1 → Apply
•Preprocessor → Element Type → Add → Add → Select Beam → 2 Node 188 → OK → Close
•Real Constants → Add → Add → Select Type Link 1 → Click OK
•Enter the cross-sectional area =1 → OK → Close
•Material Properties → Material Models → Structural → Linear → Elastic → Isotropic Enter
the Young's Modulus (EXY) = 3e7
Poisson's Ratio (PRXY) = 0.3
Sections →Beam →Common Sections →Subtype → Select Solid Circle R=0.5 N=20
T=0, Mesh view
•Preprocessor → Modeling → Create → Key points → In Active CS
•Create the key points according to the table
```

KP no	X	Y	Z
1	0	0	0
2	-12	0	0
3	12	0	0

4	0	-12	0
5	0	-12	0
6	0	12	0

```
•Modeling → Create → Lines → Lines → Straight Lines →
•Join the key points according to table
```

Line no	Join
1	1&4
2	4&5
3	5&6
4	2&5
5	3&4

```
•Preprocessor → Meshing → Mesh Attributes → All lines → Select element type Beam 188, Ok
•Meshing → Mesh tool→ set →Global 1Link1→Ok
•Lines→set→3&4 line click→2&5 line click→ok No of divisions 1→ok
• Mesh Tool→Mesh→Mesh only strut→ok Meshing → Mesh tool→ set →Global 2 Beam 188→Ok Lines→set→1&4 line click→4&5 line click→5&6 line click→ok Element egde length→1→ok
•Mesh Tool→Mesh→Mesh only Vertical line→ok Main menu→ plot Cntrls → Style → Size and Shape Click in the box against Display Element Type
```

```
2. Solution:
```

```
•Define Loads → Apply → Structural → Displacement → On key Points → Select keypoints 2 & 3→select UX,UY,UZ,ROTX,ROTY→Ok Select keypoints 2 & 3→select UX,UZ→Ok
•Modeling →Create¿Nodes→Rotate nodes CS¿By angles¿click 6th keypoint THXY →60→ok •Loads¿Apply→Structural¿Force/Moment→ click On nodes 28/Key point 6→Force/Moment value →10000
•Solution → Solve→ General Post proc→ List results→ Rection solution→ Plot results→ Defromed shape
```

Try

1. Repeat the above analysis for a landing gear subjected to compression load.

11.2 Dynamic analysis (Model analysis of Landing Gear)

A simple retractable landing gear subjected to a load of 10000 N. Find the deformation and stress developed in the landing gear.

```
1. Preferences: Structural Preprocessor:
```

```
•Preferences → Structural → H-Method → OK
•Preprocessor → Element Type → Add → Add → Select Link → 2D spar 1 → Apply
•Preprocessor → Element Type → Add → Add → Select Beam → 2 Node 188 → OK → Close
•Real Constants → Add → Add → Select Type Link 1 → Click OK
•Enter the cross sectional area =1 → OK → Close
•Material Properties → Material Models → Structural → Linear → Elastic → Isotropic Enter
the Young's Modulus (EXY) = 3e7
Poisson's Ratio (PRXY) = 0.3
Sections →Beam →Common Sections →Subtype → Select Solid Circle R=0.5 N=20
T=0, Mesh view
•Preprocessor → Modeling → Create → Key points → In Active CS
```

```
•Create the key points according to the table
```

KP no	X	Y	Z
1	0	0	0
2	-12	0	0
3	12	0	0
4	0	-12	0
5	0	-12	0
6	0	12	0

```
•Modeling → Create → Lines → Lines → Straight Lines →
•Join the key points according to table
```

Line no	Join
1	1&4
2	4&5
3	5&6
4	2&5
5	3&4

```
•Preprocessor → Meshing → Mesh Attributes → All lines → Select element type Beam 188, Ok
•Meshing → Mesh tool→ set →Global 1Link1→Ok
•Lines→set→3&4 line click→2&5 line click→ok No of divisions 1→ok
• Mesh Tool→Mesh→Mesh only strut→ok Meshing → Mesh tool→ set →Global 2 Beam 188→Ok Lines→set→1&4 line click→4&5 line click→5&6 line click→ok Element egde length→1→ok
•Mesh Tool→Mesh→Mesh only Vertical line→ok Main menu→ plot Cntrls → Style → Size and Shape Click in the box against Display Element Type
```

```
3. Solution:
```

```
• Analysis Type → New analysis → Modal → Ok
• Analysis Option → Block Lancoz → No. of Modes to extract = 5 → No. of Modes to expand = 5 → Ok
• Solve → Current LS → Ok
```

Try

1. Repeat the above analysis for a landing gear subjected to compression load.

12. Exercises on Static Analysis of Composite Structures

Static analysis of composite structures involves determining the internal forces, reactions and deflections in a beam structure under various loading conditions. This analysis helps engineers design composite structures that can safely support applied loads.

12.1 Static analysis (Carbon fiber reinforced composite)

A cantilever beam made of Carbon fiber reinforced composite is subjected to a point load as shown in Fig 3.1. Calculate the shear force, bending moment and stress distribution in the beam. L=150 cm, P=10kN, b=10cm, h=10cm

Hints

1. Preferences: Structural Preprocessor:

• Element → Add → Beam3 • Real constants → Add Set1 → Cross section area = 100 cmˆ2 → Izz = 833.33 cmˆ4 → Height = 10 cm → Ok → Close
• Material Properties → Material Models → Structural → Linear → Elastic → Isotropic → Ex = 2.5e7 → PRXY = 0.28 → Ok
• Modeling → Create → Key points → Inactive CS → (0,0,0) ;(100,0,0) ;(150,0,0) → Ok → Lines → Straight Lines → Ok
• Meshing → Size Controls → Manual Sizing → Lines → Picked Lines → No. of elements = 20 → Ok → Mesh → Lines → Ok

2. Solution:

• Analysis Type → New analysis → Static → Ok
• Define Loads → Apply → Structural → Displacement → On Key Points → Ok → All DOF Constrained → Ok → Force/Moments → On Key Points → FY = - 10000 → Ok
• Solve → Current LS → Ok

3.General Post Proc:

• Plot results → Deformed Shapes → Deformed + Un deformed

• Contour Plots → Nodal Solutions → DOF Solution → Y- Component Displacement Ok

Try

1. Repeat the above analysis for a beam made of glass fiber reinforced composite.

V. TEXT BOOKS:

1. Huei-Huang Lee, —Finite Element Simulations with ANSYS Workbench 16, SDC publications, 2 nd Edition, 2016.

VI. REFERENCE BOOKS:

2. Anderson, William J —MSC/Nastran: Interactive Training Program Wiley 1 st Edition 2015

VII. ELECTRONICS RESOURCES:

1. http://rcdata.ir/wp-content/uploads/2016/10/Mark_Drela_Flight_Vehicle_AerodynamicsBookZZ.org_.pdf
2. www.wind.civil.aau.dk/lecture/8sem_CFD/Lecture1/Lecture1.pdf
3. personalpages.manchester.ac.uk/staff/david.d.apsley/lectures/comphydr/timedep.pdf

VIII. MATERIALS ONLINE

1. Course template
2. Lab manual